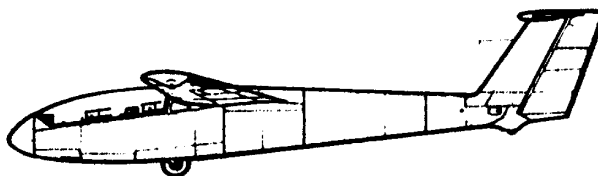




Aircraft Industries, a.s.



MANDATORY BULLETIN

MB No.: L23/052a

Concerns: L23 Super-Blaník Gliders

Subject: Check of internal elevator hinges on the stabilizer area and its potential replacement.

Reason: Appearance of cracks on the hinges.

To be accomplished by: Immediately after the bulletin receiving.

To be performed by: Non-destructive testing - Penetrant testing: Approved Organization for the testing performance according to the CSN EN 571-1 or ASTM E 165 or ASTM-E1209 or AMS-2647 or MIL-STD-271 standard requirements.
Removal, Installation, potential replacement of hinges: Approved Service Organization.

Costs to be covered by: Operator.

Necessary material to be delivered by:
Aircraft Industries a.s., Kunovice, Česká republika
against order.

Bulletin becomes effective:
On the date of its approval.

No. of pages: 3

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Ing. Pešák Miroslav
Chief Designer

Technical content of this document has been approved based on the Design Organization
Approval No. EASA.21J.119.

2. 3. 2010

A. WORK PROCEDURE

1. Preparatory works

- Remove the horizontal tail assembly of the glider according to the procedure 55-10-00, page No. 401,
- Remove the elevator from the stabilizer area according to the procedure 55-20-00, page No. 401.

2. Check of internal elevator hinges on the stabilizer area

- Checked area: Surface of the hinge web from both sides round the oval unstressing hole.
- Perform the check through non-destructive penetrant testing. Used method must be in conformance with: CSN EN 571-1 – Non-destructive testing - Penetrant testing or ASTM E-165 - Standard Practice for Liquid Penetrant Inspection or ASTM-E-1209 - Standard Method for Fluorescent Penetrant Examination Using the Water Washable Process or AMS-2647 - Fluorescent Penetrant Inspection-Aircraft and Engine Component Maintenance or MIL-STD-271 - Requirements for Non-destructive Testing. Testing parameters are depending upon the used equipment.
- Availability criteria: No cracks in checked area of the part are available.
- If no cracks are present on the checked hinges, perform record in the Airplane Logbook according to the part H, point No. 1.
- If cracks has been detected on any of the checked hinges, then perform replacement of the damaged hinge see content of the point No. 3 in this bulletin (see below).

Notice: In case some cracks were detected, send the part together with appropriate record to the producer.

The record must include information whether is the left or right hinge, the position and dimensions of cracks, the Glider SN, total flight hours and number of starts since the Glider has been put into the operation.

In case no cracks were found send the information about control execution together with the Glider SN, total flight hours and number of starts since the Glider has been put into the operation on email adress safelife@let.cz.

3. Replacement of internal elevator hinges on the stabilizer area

Notice: Perform replacement of both hinges gradually. Do not remove both hinges at the same time.

Remove rivets fixing the hinge gradually and then remove the hinge out according to the Dwg. No. A730 250N.

During removing of the rivets keep steps in the work procedure according to the MM, Chapter 20-20-02.

- Transfer holes determined for hinge grabbing to the beam web from the original hinge to the new one. Drill holes \varnothing 3.1.
- With help of strut set up new hinge according to the Dwg. No. A730 250N. Fix the hinge to the beam with helping M3 screws and nuts. Check parallelism of the hinges and alignment of the bearings.
Drill holes \varnothing 3.1 in the hinge according to the original holes in the skin.
Drill holes \varnothing 3.6. Fix the hinge with helping M3 screws and nuts.
- Install the elevator preliminary according to the procedure 055-20-00, page No. 401.
- Install the the horizontal tail assembly preliminary according to the procedure 55-10-00, page No. 401.
- Verify elevator freedom of movement
- Remove the horizontal tail assembly and the elevator.

Remove helping screws with nuts and holes Ø 3.1 redrill to fit Ø 3.6, re-drill holes to conform Ø 3.6 and then deburr the holes for rivets. Rivet the console with rivets Ø 3.5.

4. Finishing works

- Carefully clear all areas where installation work was carried out from scraps of material and from dirt.
- Install the elevator on the stabilizer area according to the procedure 55-20-00, page No. 401.
- Install the horizontal tail assembly on the glider according to the procedure 55-20-00, page No. 401.
- Perform control levelling of the stabilization area according to the MM, Chap. 8-20-00.

B. INSTALLATION KIT DELIVERED BY THE GLIDER MANUFACTURER

Referred numbers of parts are valid for replacement of one hinge only.

Replacement of both hinges requires double number of the parts.

Pos.	Identifier	Pcs/Glider	Name
1	A730 258N	1	Console
2	3531 3511	6	Rivet 3,5x11 CSN 022304.5
3	3519 3509	3	Rivet 3,5x9 CSN 022302.5
4	3519 3510	4	Rivet 3,5x10 CSN 022302.5

C. NECESSARY DOCUMENTATION

Maintenance Manual of the L23 SUPER-BLANÍK Glider
Drawing No. A730 250N

D. SPECIAL TOOLS

Strut 121-513-50002

Can be ordered at the glider manufacturer Aircraft Industries a.s., 686 04 Kunovice, Czech republic or can be produced according to the documentation ordered at the glider manufacturer.

E. SPARE PARTS IN THE OPERATION

Not influenced.

F. WEIGHT OF THE GLIDER

Not influenced.

G. RECORD IN THE AIRPLANE LOGBOOK AFTER THE BULLETIN IMPLEMENTATION

1. Check of the internal elevator hinges on the stabilizer area for potential cracks appearance has been performed according to the MB L23/052a. No cracks has been occurred.
2. Replacement of internal elevator hinge (the left one, right one, both) on the stabilizer area has been performed according to the MB L23/052a.

Date:

Performed by:.....
(legible signature of verification engineer)

H. ACCOMPANYING DOCUMENTATION

Not influenced.